

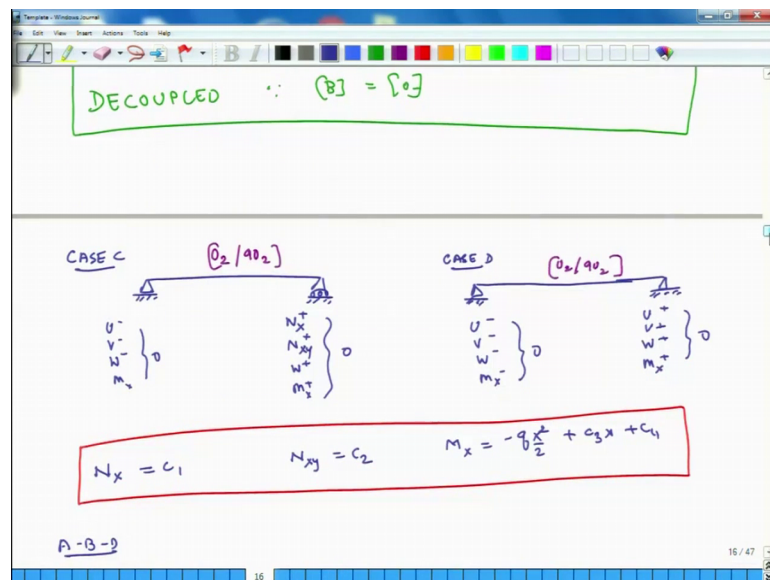
Advanced Composites
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Lecture - 35
Particular Solution for Semi-Infinite Plate
(Case C)

Hello welcome to Advanced Composites. Today is the 5th day of the on going week which is the 6th week of this course. And, what we will do today is we will start our discussion in the context of case C and case D of the Semi-Infinite Plate Problem. Now, in both these cases unlike cases A and B, the lamination sequence is not symmetric and we will see that how this influences the overall solution.

One thing based on the discussion which we had just yesterday, one thing we will very clearly see is that the in plane part of the problem and the out of plane part of the problem their solutions will be coupled. So, that is one thing we will see and then we will also see some other things which emerge out when we actually solve this problem.

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So, let us recap case C; case C is one end is pinned and fixed and the other end is pinned, but on rollers. So, here U V W and M x they are all 0 and on the other end you have N x N xy W plus and M x plus, they are all 0 and on case D both the ends are pinned as well as fixed. So, U V W M x they are 0 at both the ends and the lamination sequence is 0 2

90 2 symmetric in both the cases. So,

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I am sorry this is not symmetric it is 0 2 90 2 so, 0 2 and 90 2. So, number of layers is still the same 4 layers 2 0 layers and 2 90 layers like the earlier 2 cases A and B, but the lamination sequence is not symmetric. So, first we will very quickly solve the differential equation.

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SOLVE THE PROBLEMS TO A POINT TILL NO BC'S ARE APPLIED

$$\frac{\partial}{\partial y} = 0 \quad \frac{\partial}{\partial x} = \frac{d}{dx}$$

GOVERNING DIFF EQNS

$$\left[\begin{array}{l} \frac{\partial N_x}{\partial x} + \frac{\partial N_{xy}}{\partial y} = 0 \rightarrow \frac{dN_x}{dx} = 0 \rightarrow N_x = C_1 \quad (1) \\ \frac{\partial N_{xy}}{\partial x} + \frac{\partial N_y}{\partial y} = 0 \rightarrow \frac{dN_{xy}}{dx} = 0 \rightarrow N_{xy} = C_2 \quad (2) \\ \frac{\partial^2 M_x}{\partial x^2} + \frac{\partial^2 M_{xy}}{\partial x \partial y} + \frac{\partial^2 M_y}{\partial y^2} + \rho(x) = 0 \rightarrow \frac{d^2 M_x}{dx^2} = -q \rightarrow M_x = -\frac{q}{2}x^2 + C_3x + C_4 \quad (3) \end{array} \right.$$

The solutions for differential equation as we discussed earlier, they are going to be the same. Because, when we are doing these solutions, we do not consider the effect of lamination sequence and neither did we consider the effect of boundary condition. So, if that is the case then these 3 equations still hold. So, we will directly write these equations later. So, our 3 equations which we get from the differential equation is N_x is equal to C_1 , N_{xy} is equal to C_2 and M_x is equal to $-\frac{q}{2}x^2 + C_3x + C_4$.

So, these equations are still valid. Next we have to be now more careful when we are doing this problem. So, let us look at ABD relations, ABD matrices. Now, in this system all the layers are either 0 degrees or 90 degrees. So, it is a cross ply laminate ok, it is a cross ply laminate.

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A-B=0 Because cross ply: $\left[\begin{array}{l} A_{16} = A_{26} = B_{16} = B_{26} = D_{16} = D_{26} = 0 \\ B_{12} = B_{66} = 0 \end{array} \right]$

$$N_x = A_{11} \epsilon_x + A_{12} \epsilon_y + A_{16} \gamma_{xy} + B_{11} \kappa_x + B_{12} \kappa_y + B_{16} \kappa_{xy}$$

$$N_y = A_{11} \frac{d^2 w}{dx^2} - B_{11} \frac{d^2 w}{dx^2} = C_1$$

$$u' = \left[C_1 x + B_{11} \frac{d^2 w}{dx^2} + C_5 \right] \frac{1}{A_{11}} \quad (1)$$

$$N_{xy} = A_{16} \epsilon_x + A_{26} \epsilon_y + A_{66} \gamma_{xy} + B_{16} \kappa_x + B_{26} \kappa_y + B_{66} \kappa_{xy} = A_{66} \gamma_{xy}$$

So, when if it is a cross ply laminate because, it is a cross ply laminate we can say that it is A 1 6 is equal to A 2 6 is equal to B 1 6 is equal to B 2 6 is equal to D 1 6 is equal to D 2 6 is equal to 0 ok. And, also because this is 2 layers of 0 and 2 layers of 90 we can also show that B 1 2 is equal to B 6 6, this is also equal to 0 ok. So, if that is the case then what do we; so now, we start developing relations for N x N y N xy. So, what is it N x is equal to A 1 1 epsilon x dot plus A 1 2 epsilon y dot plus A 1 6 gamma xy at the mid plane. A 1 6 is 0 oh I am sorry, but they will also B terms plus B 1 1 K x plus B 1 2 K y plus B 1 6 K xy. So, now, things start becoming messy.

So, now let us see which terms can be dropped off, A 1 6 is, 0 B 1 6 is also 0 epsilon y is 0 because del over del y is 0. So, this is 0 K y is 0 because, again it is related to partial second derivative of w in the y direction and K xy is also 0 because, K xy is del 2 w over del x del y. So, what this gives us is equal to A 1 1 d naught over d x minus B 1 1 d 2 w naught over dx square N x. So, now you see the coupling very explicitly N x is a function of u and w. In the symmetric laminate this B term would have been 0. So, N x would only depend on u, but here N x depends on u as well as on w. So, it is coupled problem and we know that N x is equal to C 1 from this red box N x is equal to C 1.

So, we integrate this equation once and we get U naught is equal to C 1 x plus B 1 1 dw naught over d x plus C 5 into 1 over A 1 1 ok. So, we will call this equation 1. Now, let us look at the relation for N xy. So, N xy is equal to A 1 6 epsilon x naught plus A 2 6 epsilon y naught plus A 6 6 gamma xy naught plus B 1 6 K x plus B 2 6 K y plus B 6 6 K xy and let us see which terms get dropout in this equation. So, A 1 6 is 0 A 2 6 is 0

because, it is a cross ply laminate; B_{16} is 0 B_{26} is 0 and B_{66} is also 0. We have mentioned this and actually also K_{xy} is 0 ϵ_y is 0 so, all these things are 0. So, essentially this boils down to $A_{66} \gamma_{xy}$ everything else goes away.

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$$N_x = A_{11} \epsilon_x^0 + A_{12} \epsilon_y^0 + A_{16} \gamma_{xy}^0 + B_{11} \kappa_x^0 + B_{12} \kappa_y^0 + B_{16} \kappa_{xy}^0$$

$$\left. \begin{aligned} \epsilon_x^0 &= \frac{\partial u^0}{\partial x} \\ \epsilon_y^0 &= \frac{\partial v^0}{\partial y} \\ \gamma_{xy}^0 &= \frac{\partial u^0}{\partial y} + \frac{\partial v^0}{\partial x} \end{aligned} \right\}$$

N_x only depends on $\epsilon_x^0, \epsilon_y^0, \gamma_{xy}^0$.
 N_y " " " " " "
 N_{xy} " " " " " "

$(N_x, N_y, N_{xy}) \rightarrow u^0 \text{ and } v^0$

Solution of inplane problem only involves u^0 and v^0 .

And γ_{xy} if you remember somewhere back we had said γ_{xy} is $\frac{\partial u}{\partial y} + \frac{\partial v}{\partial x}$, but $\frac{\partial u}{\partial y}$ is 0. So, it is $\frac{\partial v}{\partial x}$ only.

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$$N_{xy} = A_{66} \frac{dv^0}{dx} = c_2$$

$$v^0 = \left[c_2 x + c_3 \right] \frac{1}{A_{66}} \quad \textcircled{1}$$

$$N_{xy} = A_{16} \epsilon_x^0 + A_{26} \epsilon_y^0 + A_{66} \gamma_{xy}^0 + B_{16} \kappa_x^0 + B_{26} \kappa_y^0 + B_{66} \kappa_{xy}^0 = A_{66} \gamma_{xy}^0$$

$$N_{xy} = A_{66} \frac{dv^0}{dx} = c_2 \rightarrow v^0(x) = \left[c_2 x + c_3 \right] \frac{1}{A_{66}} \quad \textcircled{2}$$

$$M_x = B_{11} \epsilon_x^0 + B_{12} \epsilon_y^0 + B_{16} \gamma_{xy}^0 + D_{11} \kappa_x^0 + D_{12} \kappa_y^0 + D_{16} \kappa_{xy}^0$$

$$= B_{11} \epsilon_x^0 + D_{11} \kappa_x^0 = B_{11} \frac{du^0}{dx} - D_{11} \frac{d^2 u^0}{dx^2} = -\frac{q_0^2}{2} + c_4 x + c_5$$

$$B_{11} v^0 - D_{11} \frac{d^2 v^0}{dx^2} = -\frac{q_0^2}{6} + c_4 x + c_5 \quad \textcircled{3}$$

So, we can say that N_{xy} is equal to $A_{66} \frac{dv^0}{dx}$. So, this gives us v^0 I am sorry v^0 which is the function of x is equal to and what is N_{xy} ; N_{xy} is c_2 we

have already said that earlier right from this equation $N_{xy} = C_2$. So, this is equal to $C_2 x$ plus C_6 into 1 over A_{66} . So, this is the expression for V , this is the expression for V . Next look at M_x ; so, M_x equals $B_{11} \epsilon_x$ plus $B_{12} \epsilon_y$ plus $B_{16} \gamma_{xy}$ plus $D_{11} K_x$ plus $D_{12} K_y$ plus $D_{16} K_{xy}$ ok.

Again we see which terms can be dropped out B_{16} is 0 D_{16} is 0 and K_y is 0 . So, what I am left with is $B_{11} \epsilon_x$ plus oh actually ϵ_y is also 0 . So, $B_{11} \epsilon_x$ plus $D_{12} K_x$ that is all I am left and this is equal to I am sorry this is should be D_{11} . So, this is equal to $B_{11} \frac{du}{dx}$ minus $D_{11} \frac{d^2 w}{dx^2}$ ok. Now, this is equal to M_x , but we also have another expression for M_x which is this one. M_x is equal to minus $q x^2$ plus $C_3 x$ plus C_4 which is valid for all the 4 cases so, I putting in there; so, minus $q x^2$ over 2 plus $C_3 x$ plus C_4 .

So, if I integrate this whole thing ones and I rearrange my things what I get is $B_{11} U$ naught minus $D_{11} \frac{dw}{dx}$ over dx is equal to minus $q x^3$ over 6 plus $C_3 x^2$ over 2 plus $C_4 x$ plus C_7 . So, this is my third equation, this is the third equation. So, let us quickly see now, we have expressions for U V and W . So, what we see is that equation 1 2 and 3 U involves dw over dx so, it is a coupled problem. V does not involve any other thing and the third equation is $B_{11} U$ naught minus $D_{11} \frac{dw}{dx}$ over dx . So, again there is a coupling between U and W in the third equation ok.

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$$N_{xy} = A_{66} \frac{d\gamma^0}{dx} = C_2 \rightarrow v^0(x) = [C_2 x + C_6] \times \frac{1}{A_{66}} \quad (2)$$

$$M_x = B_{11} \epsilon_x^0 + B_{12} \epsilon_y^0 + B_{16} \gamma_{xy}^0 + D_{11} K_x^0 + D_{12} K_y^0 + D_{16} K_{xy}^0$$

$$= B_{11} \epsilon_x^0 + D_{11} K_x^0 = B_{11} \frac{du^0}{dx} - D_{11} \frac{d^2 w^0}{dx^2} = -\frac{qx^2}{2} + C_3 x + C_4$$

$$B_{11} v^0 - D_{11} \frac{dw^0}{dx} = -\frac{qx^2}{6} + C_3 \frac{x^2}{2} + C_4 x + C_7 \quad (3)$$

Use (1) and (3) to solve for $v^0(x)$ and $\frac{dw^0}{dx}$.
 Integrate the relation for dw^0/dx to get eqn. for $w^0(x)$.

So, what we can do is we can solve. So, what we can do is we can use 1 and 3 to solve

for U naught as a function of x and dw naught over $d x$ as a function of x . These are simultaneous equations so, I can solve for U naught and dw over dx ok. So, I will get one expression for U naught and another expression for $d w$ over $d x$ and then integrate the relation for dw over $d x$ to get equation for W naught as a function of x ok. So, if I do all the math; so, you can do all that math that fairly standard 12th standard mathematics of linear simultaneous equation. So, if we do what we end up with is a relation which looks something like this, I will write it.

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The image shows a handwritten slide with the following equations and definitions:

$$w^{\circ}(x) = \frac{1}{D_{11}} \left[\frac{q x^4}{24} - c_2 \frac{x^3}{6} + \left(\frac{B_{11}}{A_{11}} \cdot c_1 - c_4 \right) \frac{x^2}{2} + \left(\frac{B_{11}}{A_{11}} c_5 - c_7 \right) x + c_8 \right]$$

$$u^{\circ}(x) = \frac{1}{A_{11}} \left[\frac{B_{11}}{A_{11}} \frac{q x^3}{6} - \frac{B_{11}}{A_{11}} \frac{c_2 x^2}{2} + \left(c_1 - \frac{B_{11}}{A_{11}} c_4 \right) x + \left(c_5 - \frac{B_{11}}{A_{11}} c_7 \right) \right]$$

$$v^{\circ}(x) = (c_2 x + c_6) \times \frac{1}{A_{66}}$$

Definitions:

$$D_{11} = D_{11} - \frac{B_{11}^2}{A_{11}} \quad A_{11} = A_{11} - \frac{B_{11}^2}{D_{11}}$$

Reduced Bending Stiffness:

$$M_x = -q \frac{x^2}{2} + c_3 x + c_4 \quad M_y = -D_{12} \frac{d^2 v}{dx^2} \quad M_{xy} = 0 \quad \therefore B_{12} = 0$$

CASE C & D

So, W naught x is equal to 1 over D_{11} . Now, this is not regular D this is a difference symbol D . So, it is it is not regular. So, I will define what this big D_{11} is ok. It is not like our regular D times $q x^4$ over 24 minus $C_3 x^3$ over 6 plus B_{11} over A_{11} times $C_1 C_4 x^2$ over 2 plus B_{11} over A_{11} times C_5 minus C_7 times x plus C_8 ok. And the relation for U naught x is equal to 1 over A_{11} , again this is a different A it is not like our regular A . And, this times B_{11} over A_{11} times $q x^3$ over 6 minus B_{11} over A_{11} times $C_2 x^2$ over 2 plus C_1 minus B_{11} over A_{11} times C_4 plus B_{11} over A_{11} times C_7 actually it better. So, it will be C_5 minus B_{11} over A_{11} times C_7 and then of course V .

What is V naught? It is equal to $C_2 x$ plus C_6 into 1 over A_{66} and I have defined these you know script D_{11} and script A_{11} . So, what are these? This D_{11} equals regular D_{11} minus B_{11} square over A_{11} and this is script A_{11} equals A_{11} minus B_{11} over D_{11} square and so, these are U , V and W 's. And then what is M_x , is still minus $q x^2$

over 2 plus $C_3 x$ plus C_4 . M_y is still minus $D_{12} D_2 w$ naught over $d x$ square, M_{xy} is still 0 because, B_{12} is 0. So, this is the solution for both cases. So, this is the solution for case C and case D because, we have not implemented the boundary conditions still so far. So, this is the solution which is common 2 cases C and D.

Once again this solution has a lot of unknowns $C_1 C_2 C_3 C_4 C_5 C_6 C_7$ and C_8 and those 8 unknowns will be computed once we start implementing the 8 different boundary conditions. 2 more points, this term this D_{11} script D_{11} this is called Reduced Bending Stiffness ok; it is called reduced bending stiffness. Why is it reduced bending stiffness? Because, D_{11} by itself is positive and B_{11} is so, the effective bending stiffness in the expression for W , in the expression for W we have 1 over this script D_{11} . If B_{11} was not there then this is script D_{11} would be same as D_{11} , but because this B_{11} is there its effective value becomes reduced.

So, the denominator instead of D_{11} becomes script D_{11} and that is a smaller number; when it is a smaller number the value of W will increase. So, what this shows? This relation even though we have yet not implemented the boundary conditions; what it shows is that the presence of D matrix tends to increase the deflection of the plate. In the symmetric laminate this B was 0. So, the deflection would be less because, it will be just D_{11} . In unsymmetric laminate because B_{11} is non-zero, it reduces the stiffness. When it reduces the stiffness happens this plate bends more ok.

Same thing also happens to script A_{11} , its value is A_{11} minus B_{11} square by D_{11} . So, here also the value of U also tends to become larger because of presence of B matrix. So, this is what I wanted to discuss today. Tomorrow we will start implementing the boundary conditions and see what the actual solutions look like.

Thank you.